

# REPORT DOCUMENTATION PAGE

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DTBV 2308 m13c

FROM: PROI (STINFO)

10 May 2002

SUBJECT: Authorization for Release of Technical Information, Control Number: AFRL-PR-ED-AB-2002-109  
Ed Coy, Doug Talley, Jonathan Watts, "Pulse Combustion Rockets for Space Propulsion Applications"

5219

41<sup>st</sup> AIAA Aerospace Sciences Meeting & Exhibit  
(Reno, NV 6-9 Jan 03) (Deadline: 06 ~~May~~ 2002)

(Statement A)

Jun

Abstract for 41<sup>st</sup> AIAA Aerospace Sciences Meeting

## Pulse Combustion Rockets for Space Propulsion Applications

Edward B. Coy  
Douglas G. Talley  
Jonathan M. Watts

Air Force Research Laboratory  
Edwards AFB, CA

Pulse combustion propulsion devices are currently being considered as alternatives to conventional constant-pressure engines. Potential advantages include reduction or elimination of pumps and/or compressors, and improved Isp for a given feed system supply pressure. In this paper we compare pulse combustors, or constant-volume engines, with pulse detonation engines and discuss why in some applications the former may be the preferred cycle. A model is presented for a monopropellant-fueled, constant-volume, pulse combustor which includes finite-rate processes for injection, heat release and exhaust. The model is used to explore the time and dimensional scales of the device and to predict performance and optimal geometry. The pulsed propulsion device is found to have nearly identical specific impulse as the steady-state engine operating with the same mass flow and throat area, and the nozzle optimizes at the same area ratio. Pulsed combustor behavior is found to depend on two time scales: the ratio of the heat release time to the chamber blow-down time, and the ratio of the blow-down time to the injector pulsing period. We briefly consider the application of pulse combustion devices in pressure-fed satellite propulsion systems and examine the effect on satellite mission.

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